



Topology Optimization for Thermo-mechanics

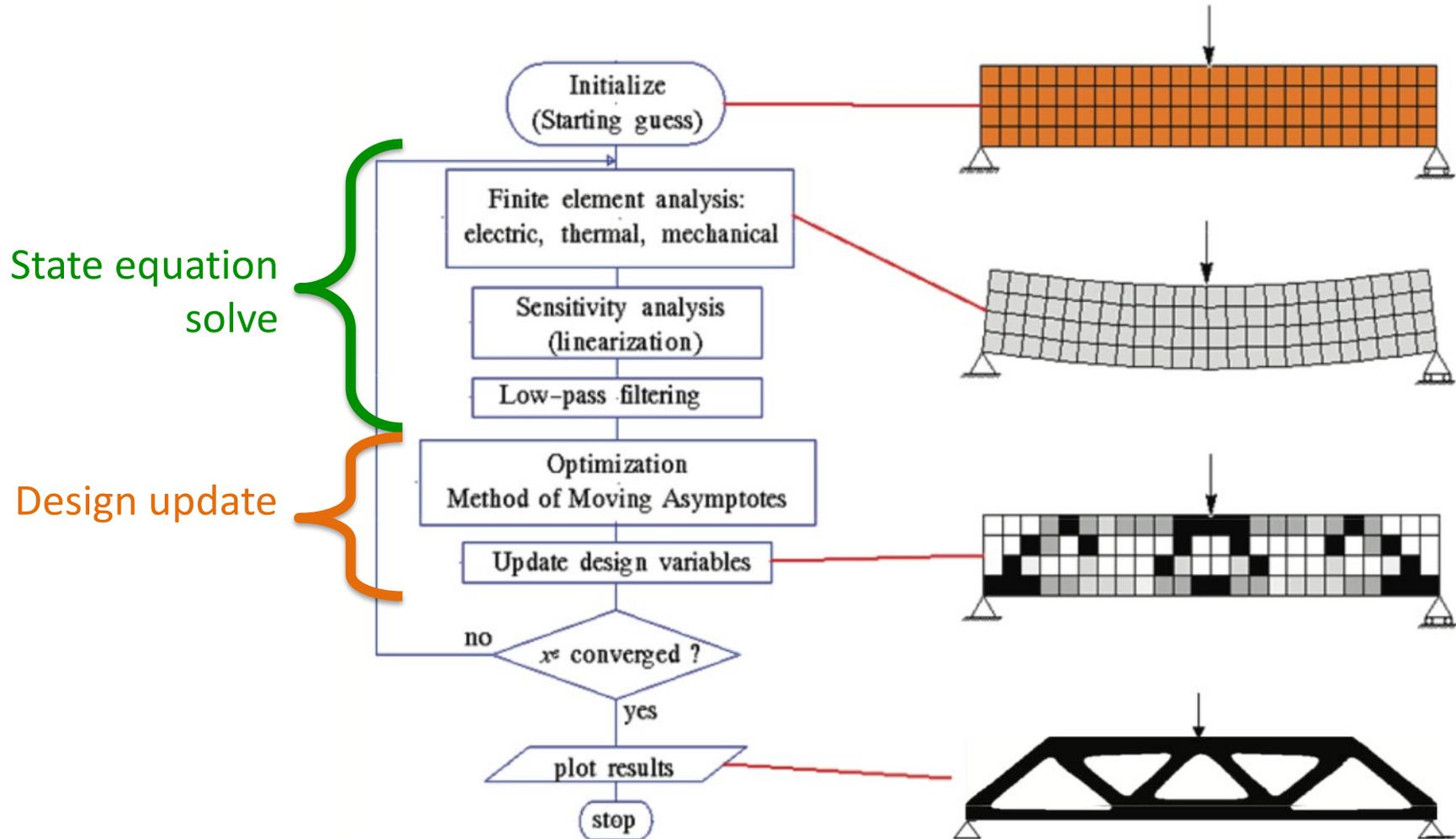
H Alicia Kim
Jacobs Scholar Chair Professor
M2DO Lab
UC San Diego



Topology Optimization

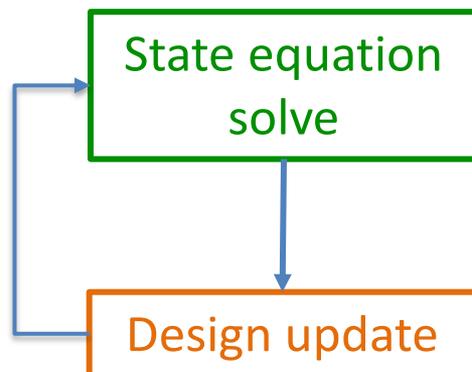


Topology Optimization



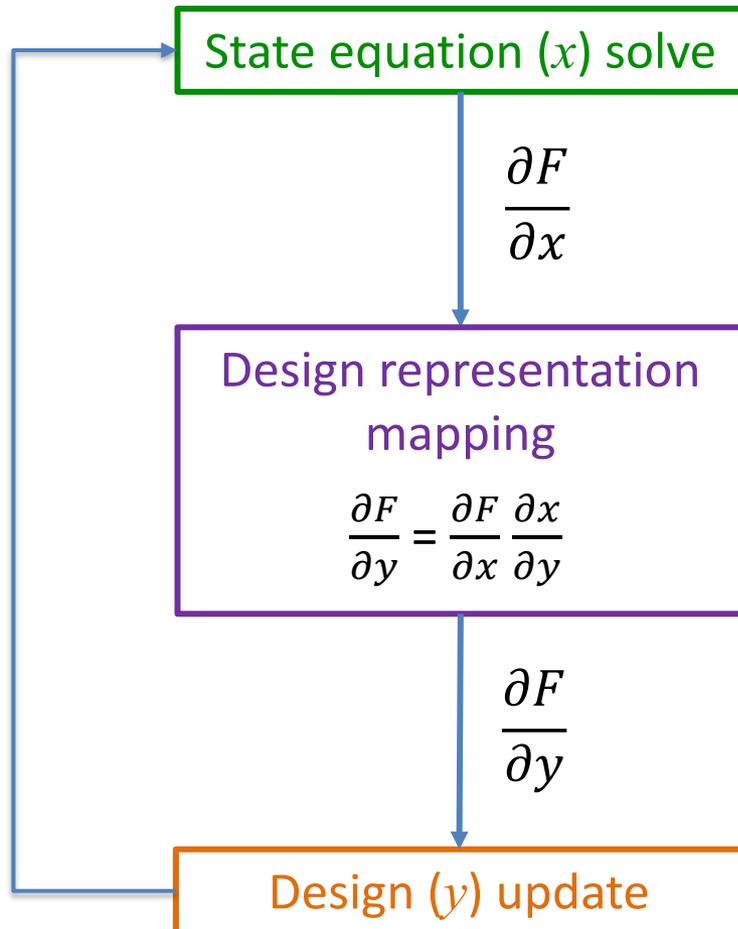


Topology Optimization





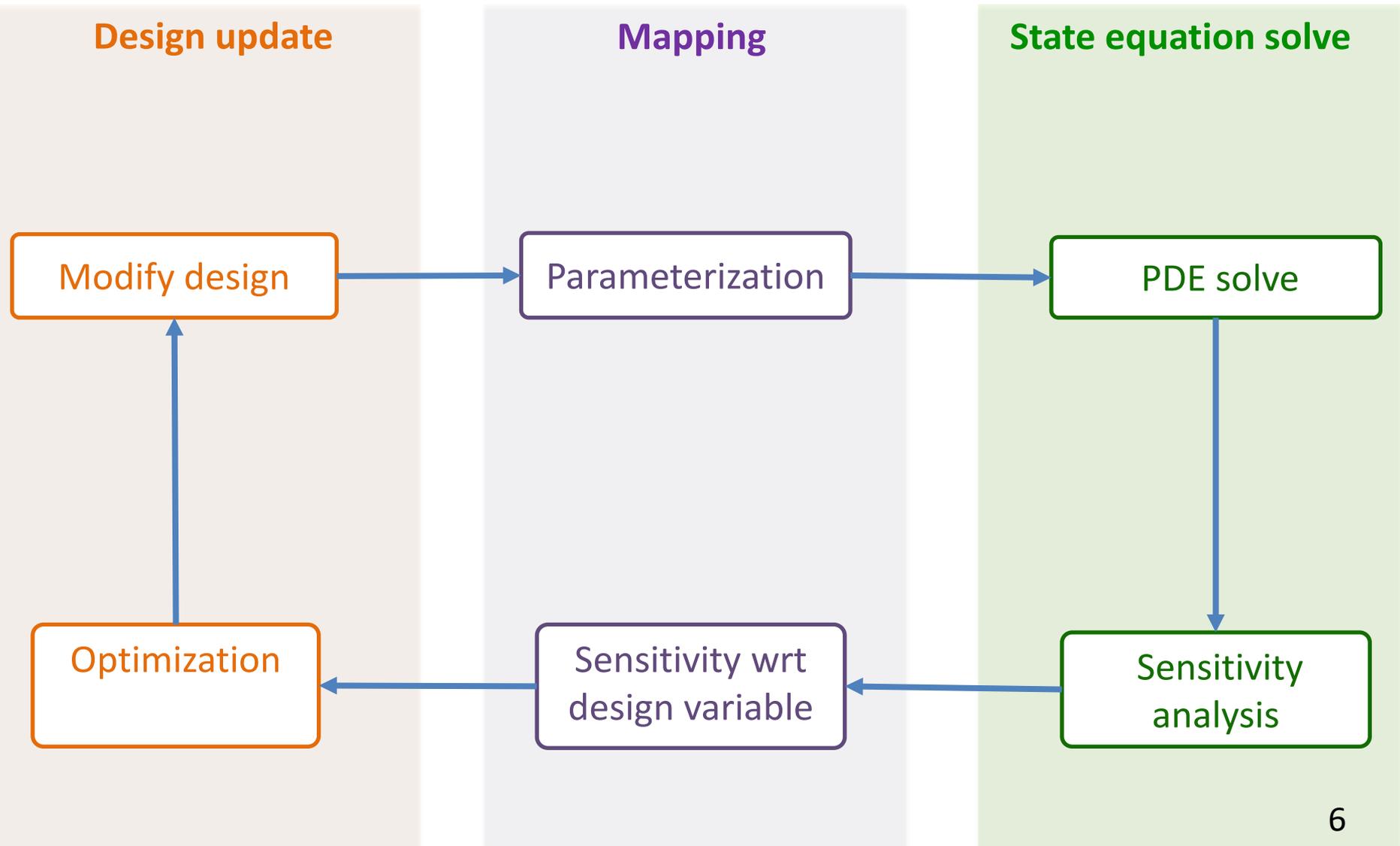
Topology Optimization



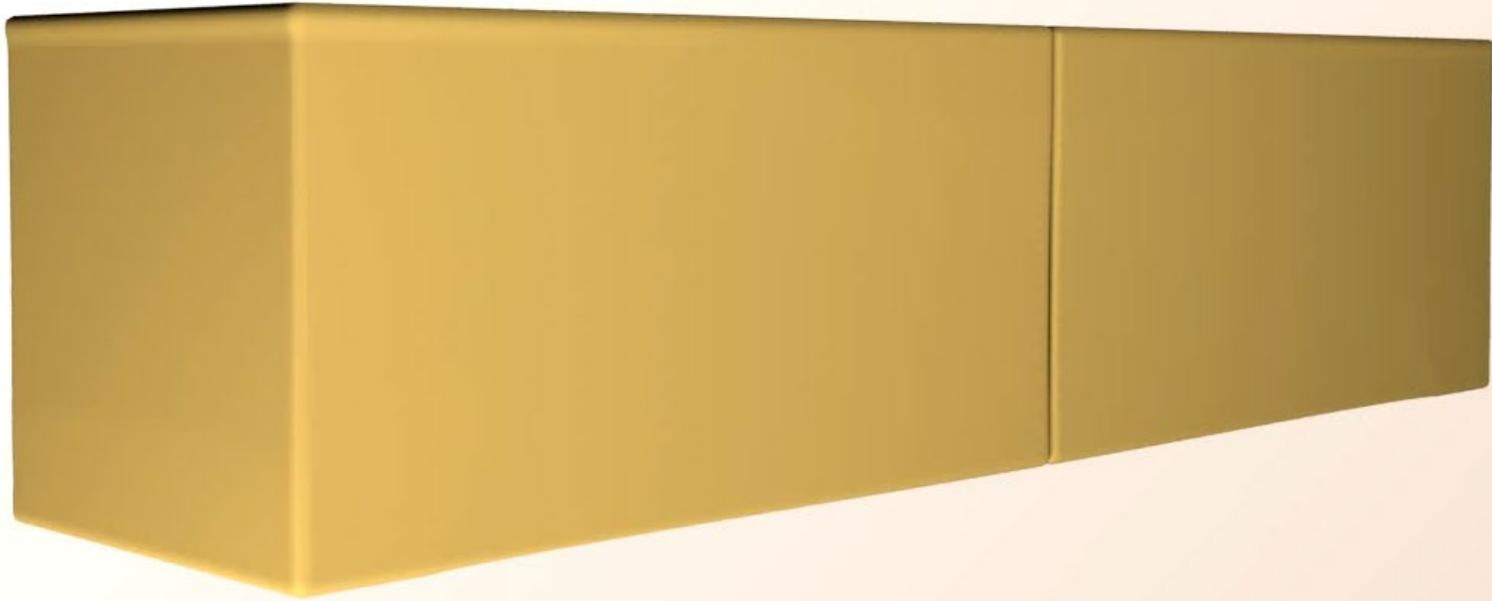
- Any PDE solver with adjoint sensitivity can be used.
- Any design parameterization can be used.



Generalized Architecture



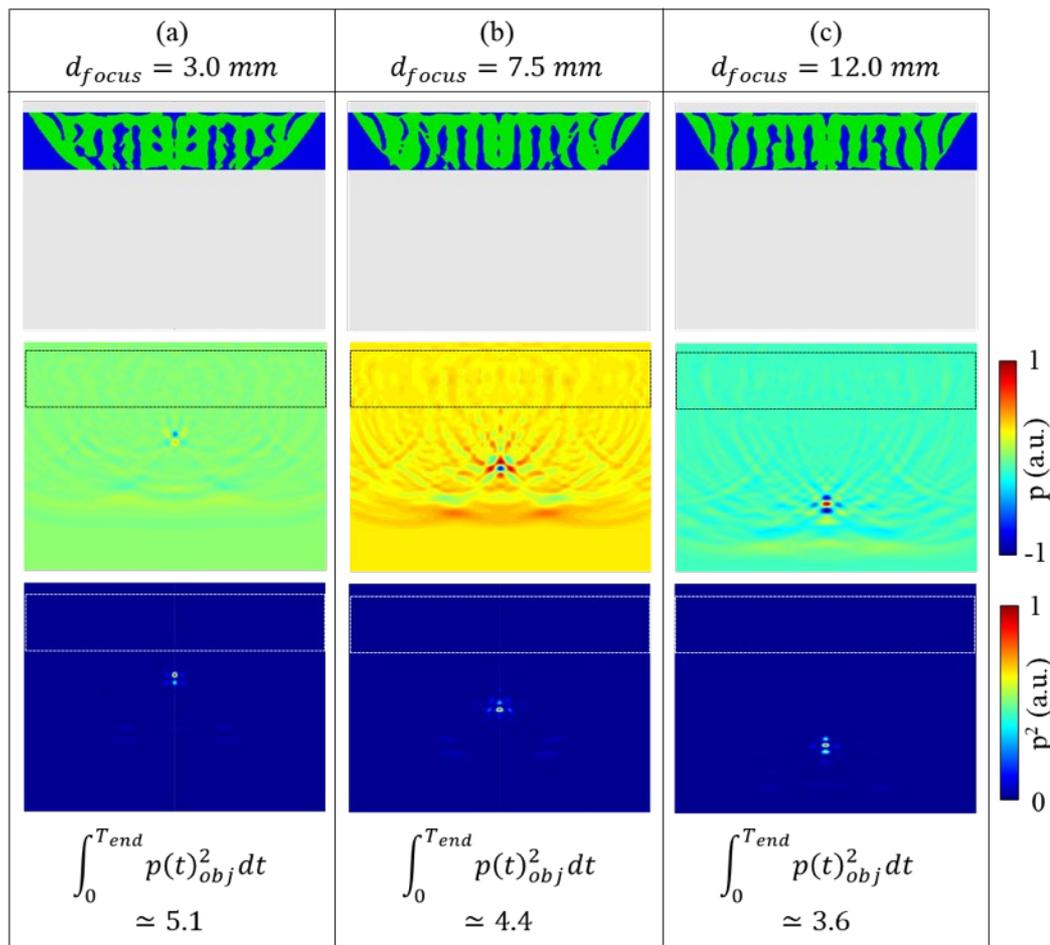
Large-Scale Level Set Topology Optimization



Modeled only a half (symmetry)
8 million elements
160 processors (64 GB RAM/node)
Converges in 250 iterations, 6 hours

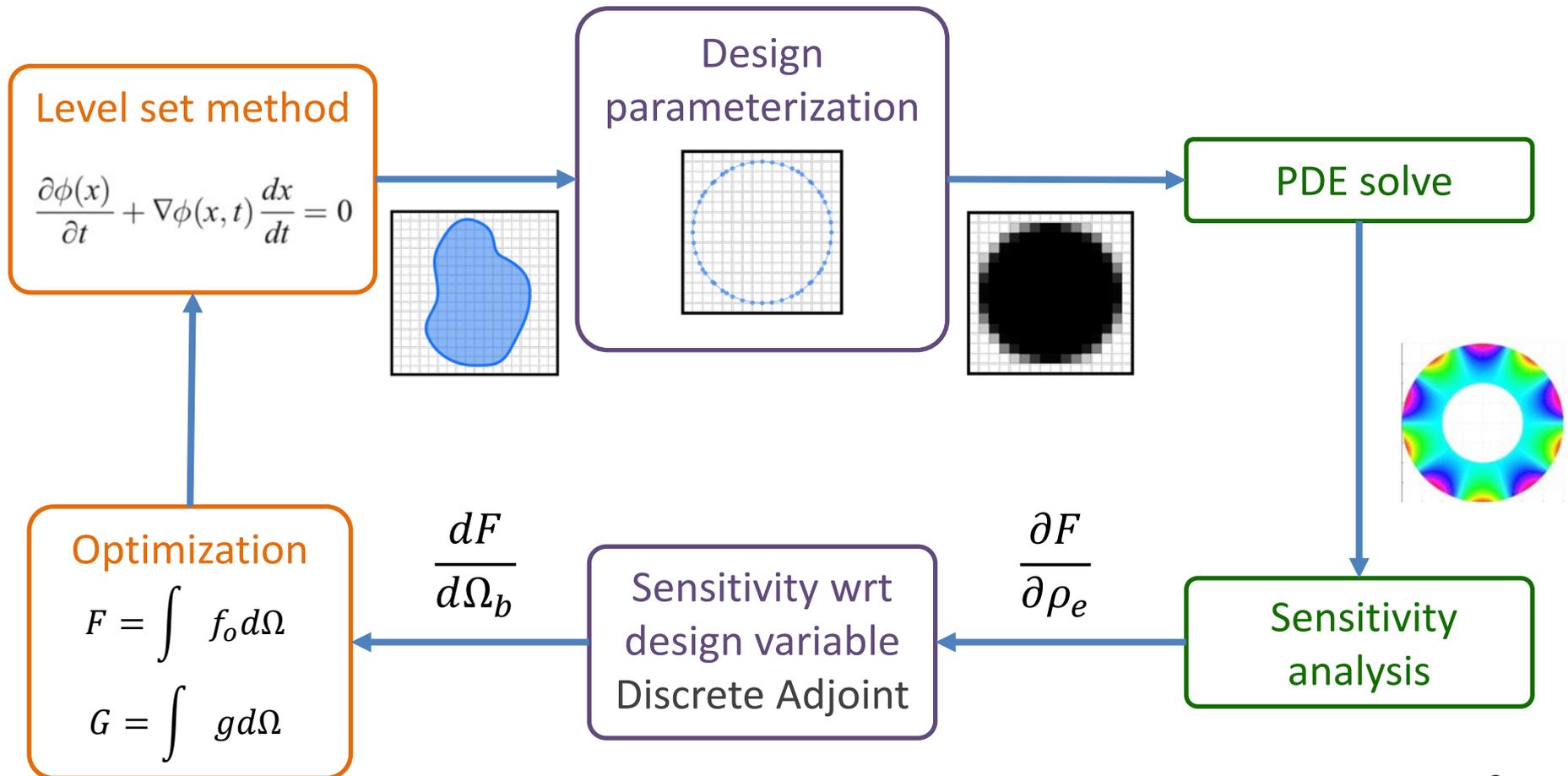


Flat Lens via Multiple Material



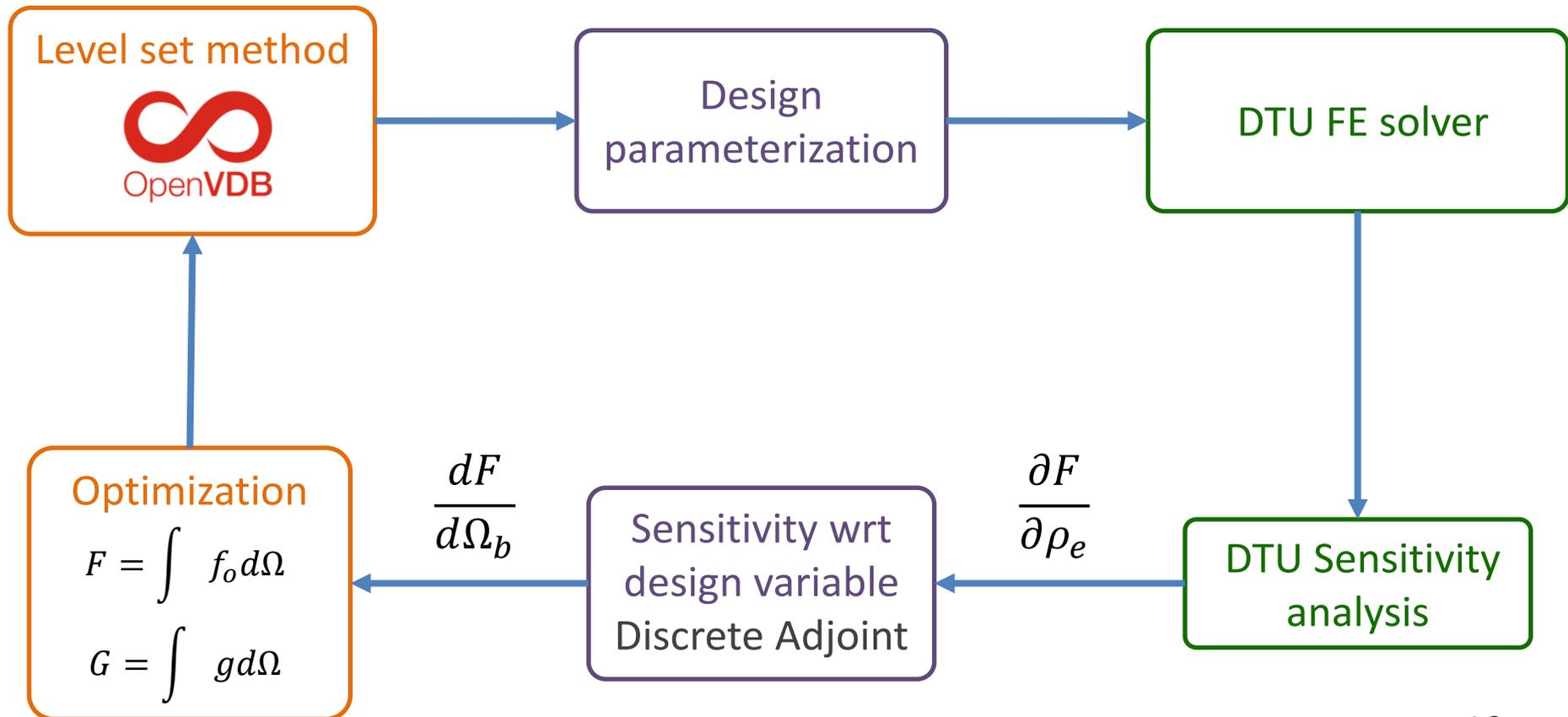
Acoustic lens design with varying focal depths

Level Set Topology Optimization



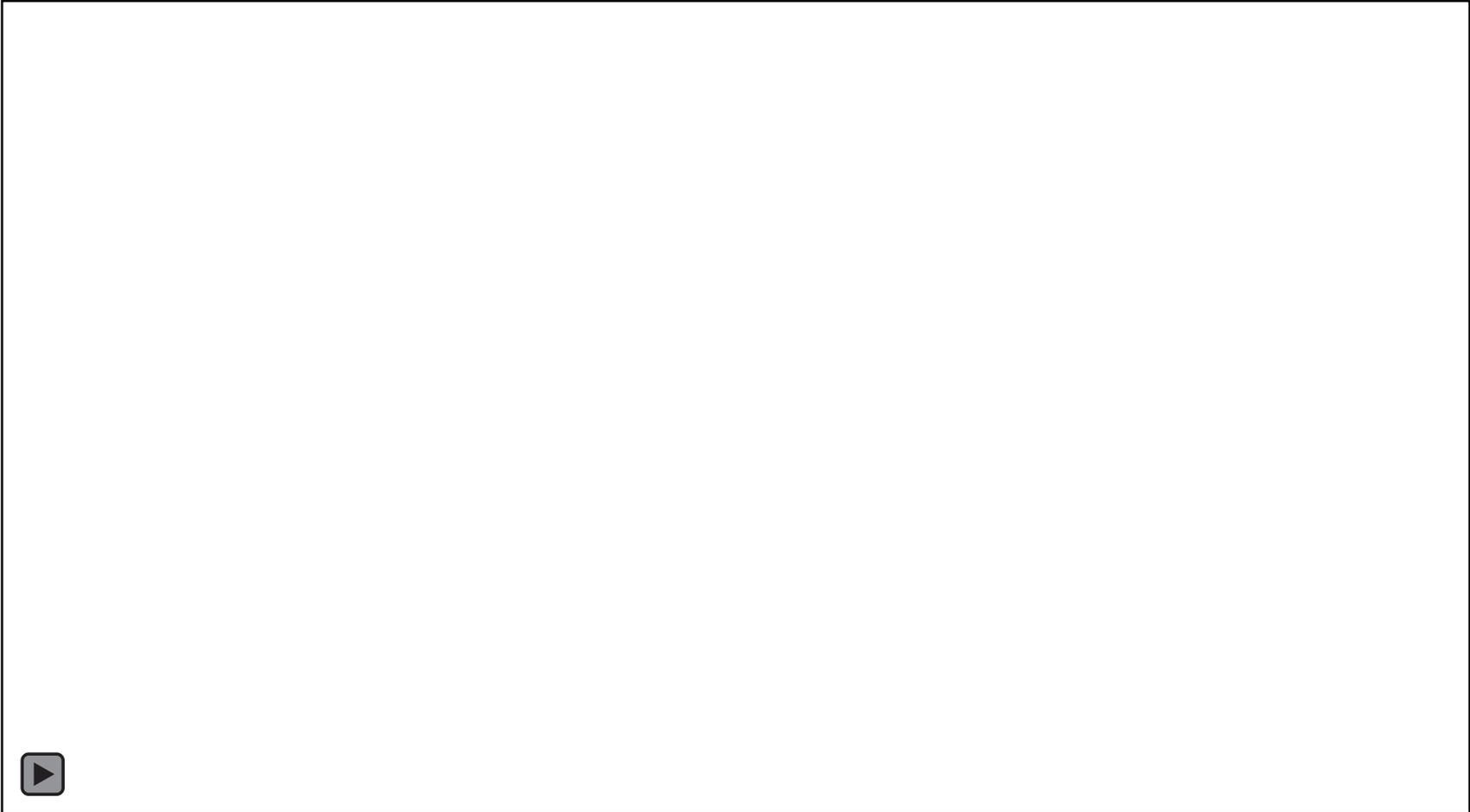


Large-Scale Topology Optimization



VDB-LSTO Method

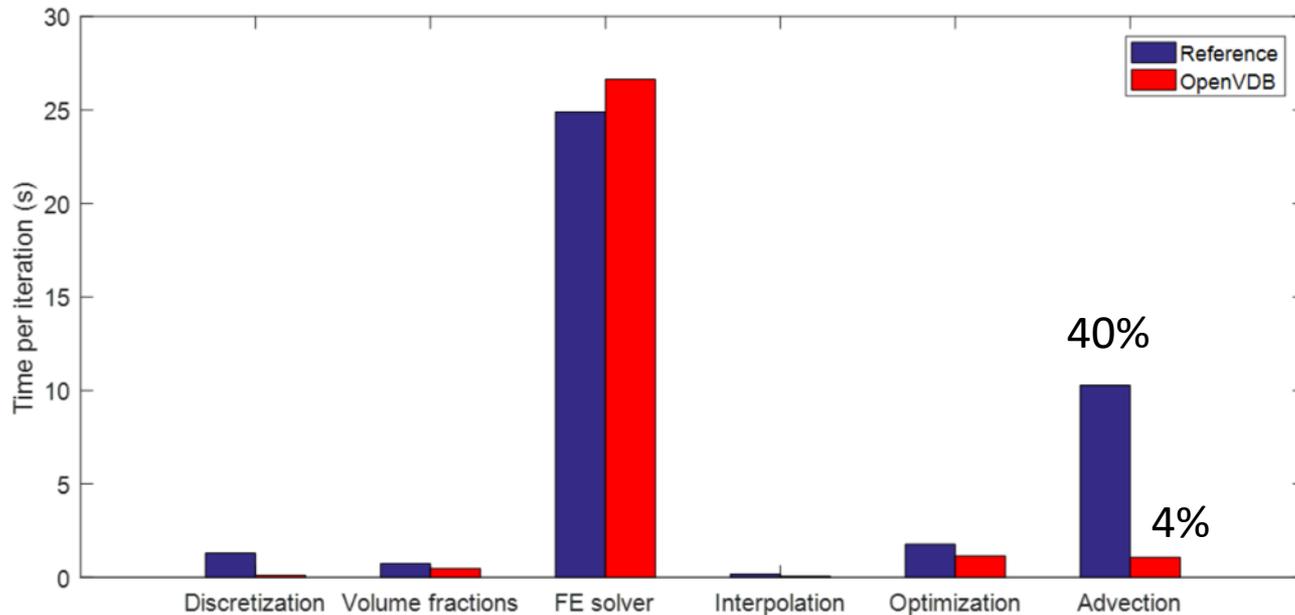
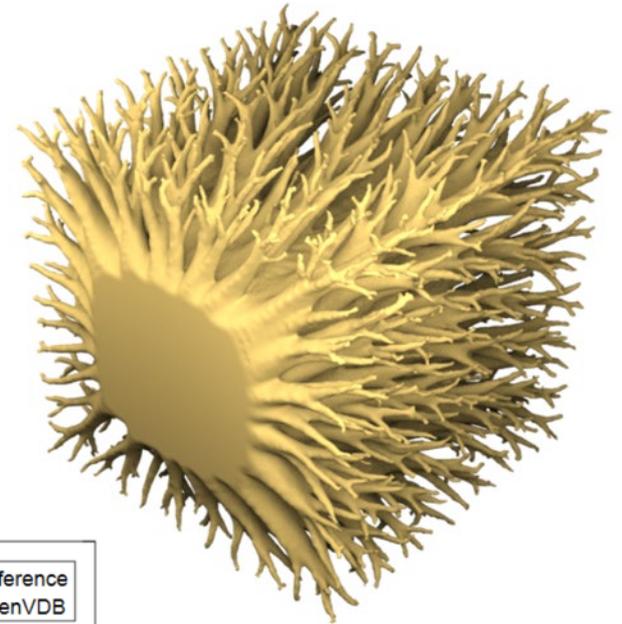
[3.2 billion elements, 56 cores, 128GB]





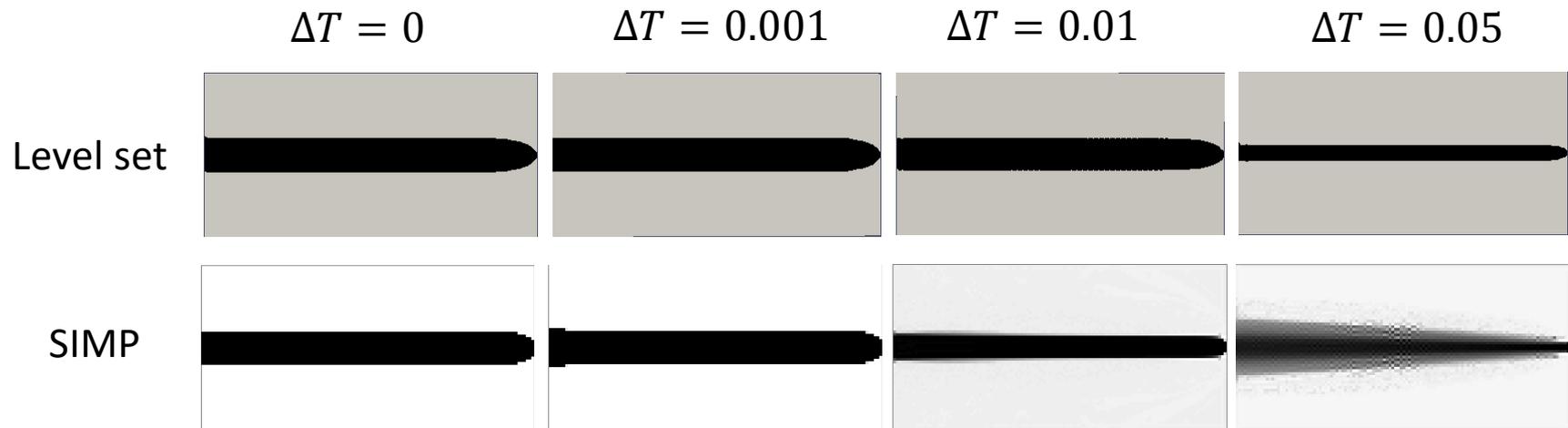
Heat conduction

- Min average temperature, st vol < 30%
- 8 million elements
- Desktop workstation, 56 cores ,128 GB RAM
- Converges in 200 iterations



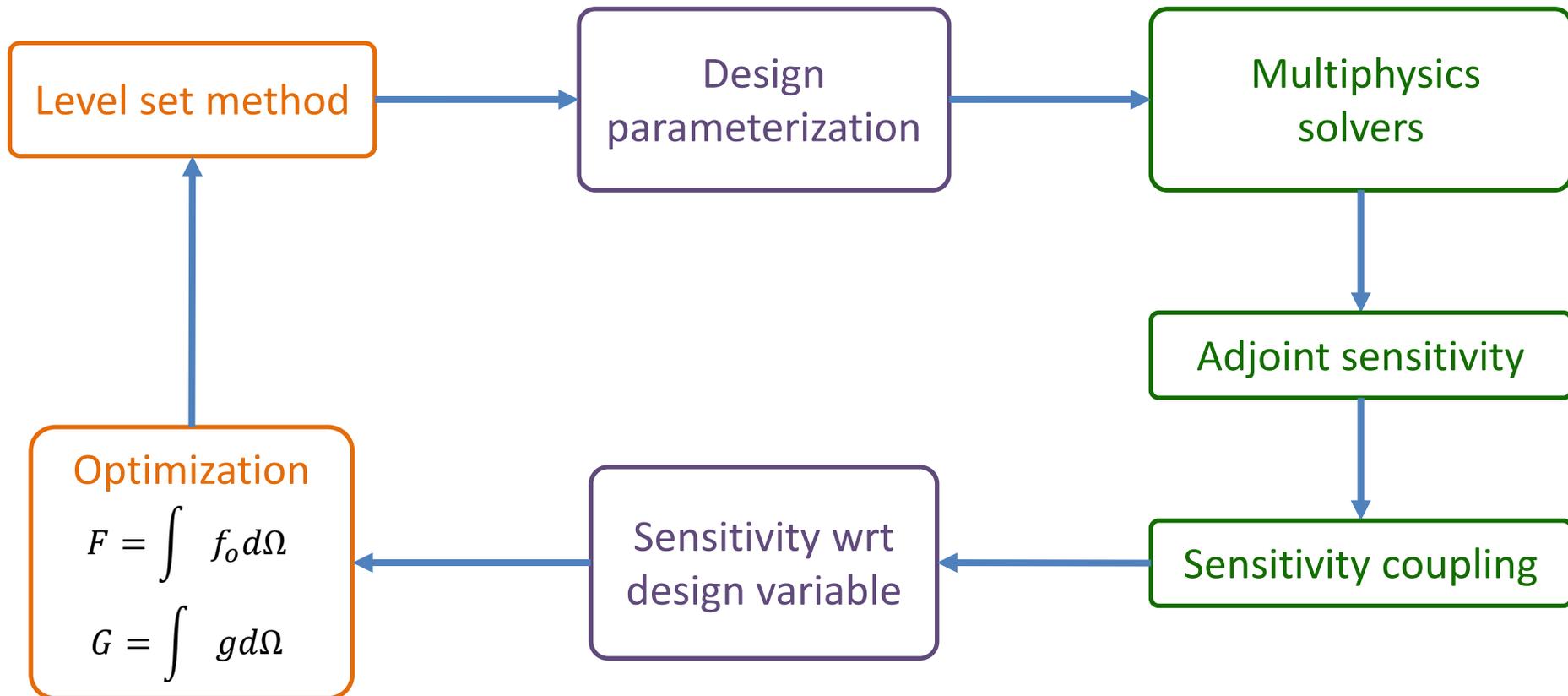


Why use Level Set Topology Optimization?





Generalized TO: Interoperable Reusable Software Architecture





ParaLeSTO: FAIR Principles for Opensource Software

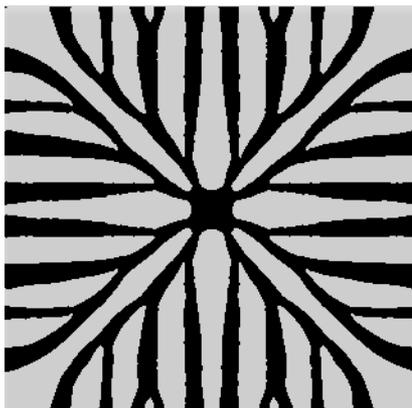
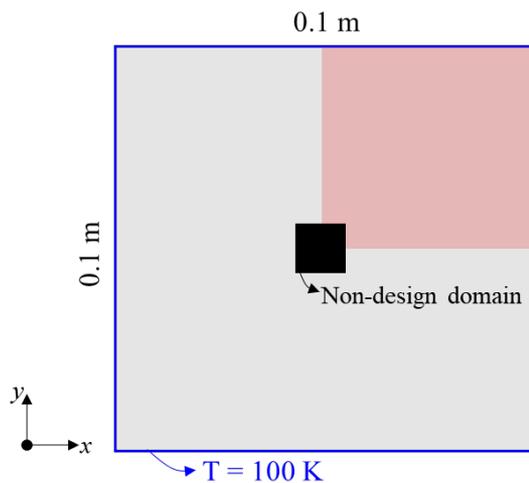
- Findability
- Accessibility
- Interoperability
- Reusability

More information at <http://m2do.ucsd.edu/software/>



Maximizing Heat Conduction via Eigenvalue

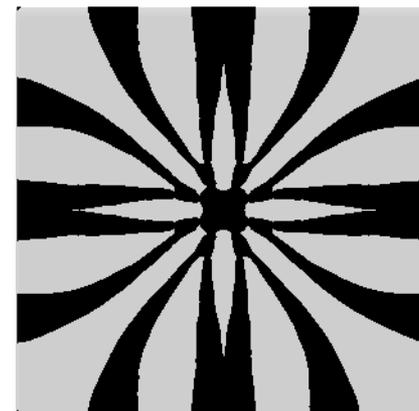
- COMSOL (Physics solver) & Symbolic AD
- Max. 1st thermal eigenvalue s.t. Locally-averaged volume



$$R_{loc} = 8$$

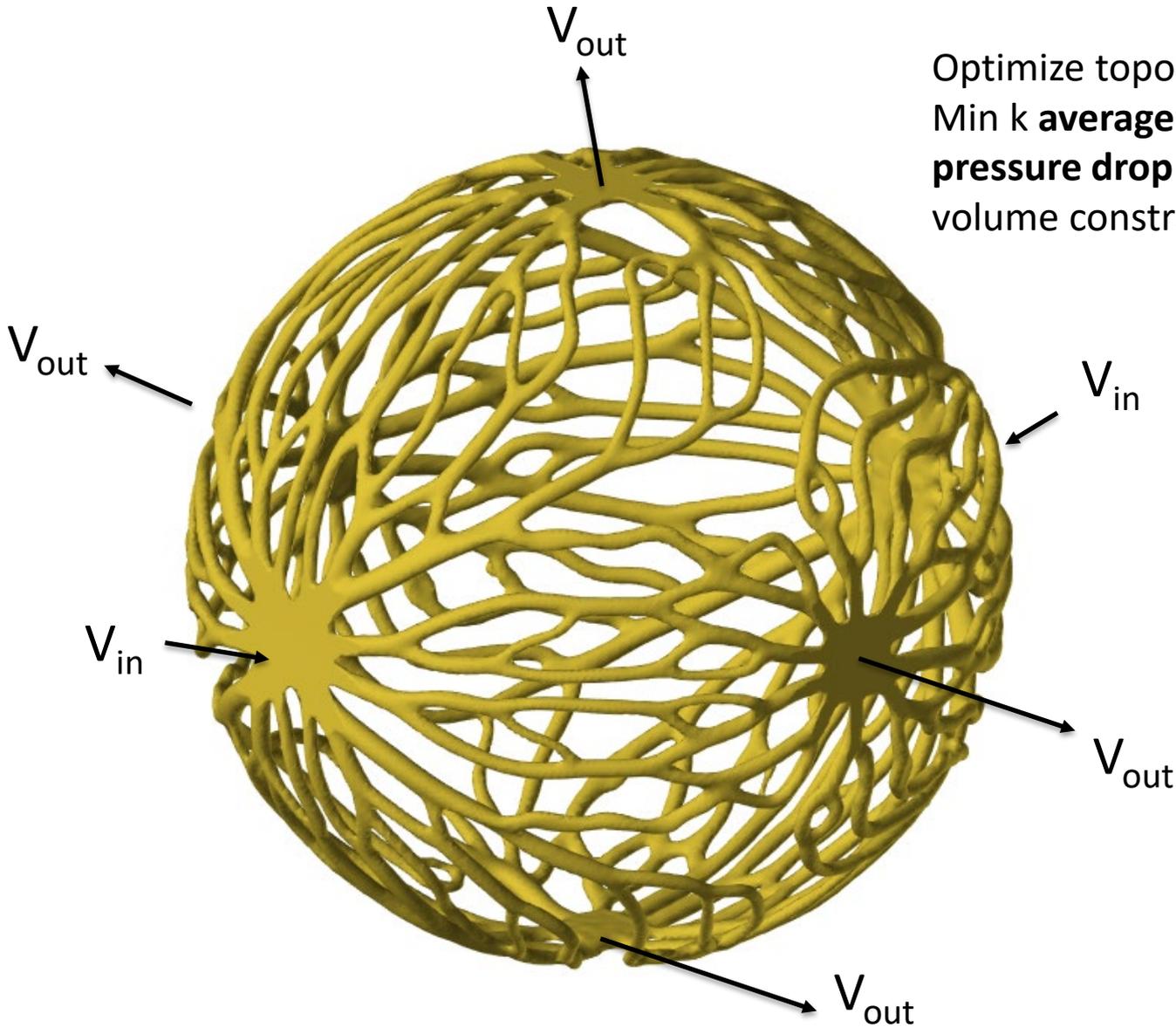
<

Manufacturable



$$R_{loc} = 14$$

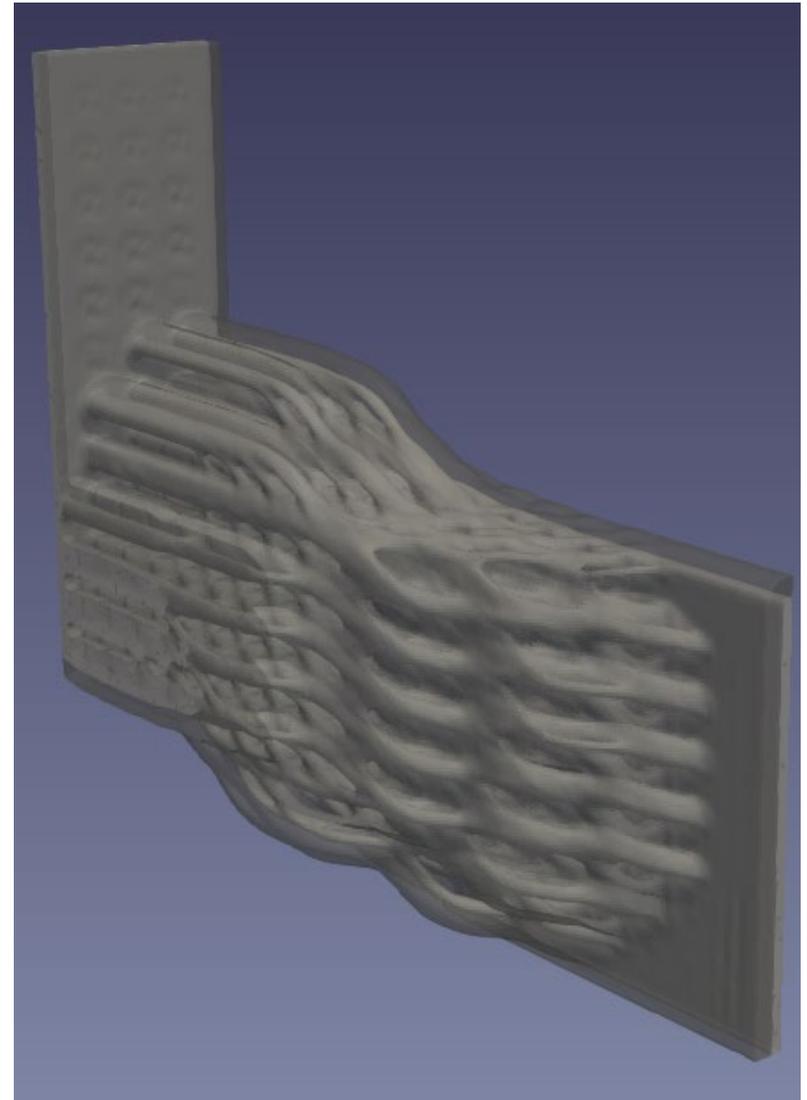
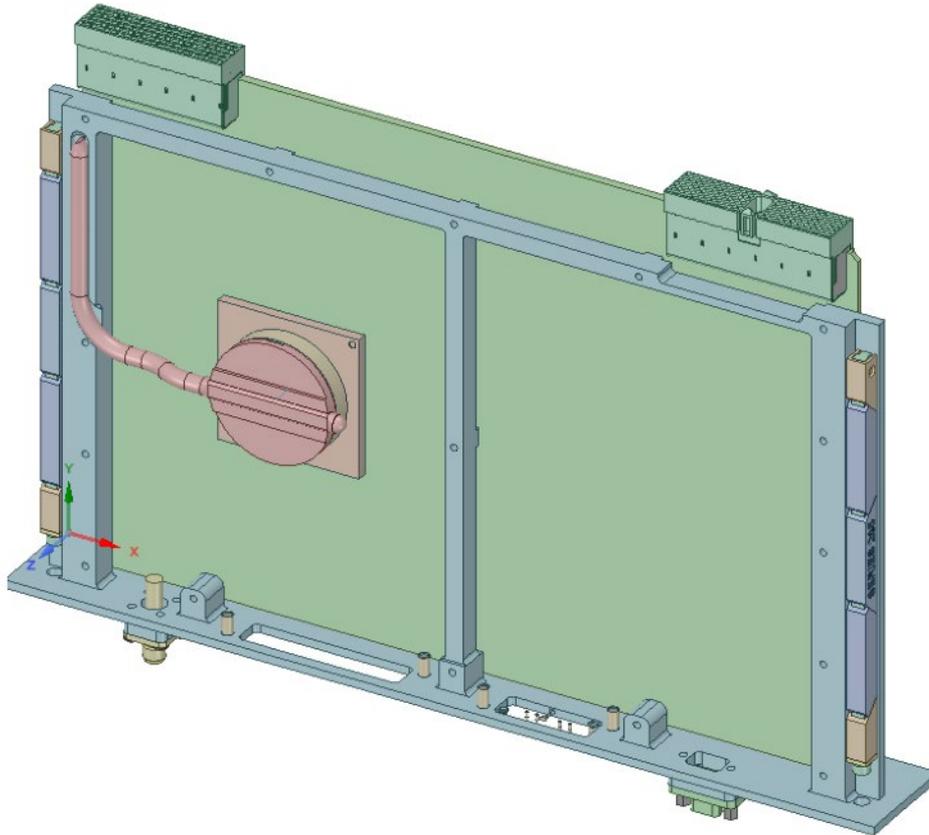
Conformal cooling channel on a sphere



Optimize topology of channel flow
Min k average temperature + $(1-k)$
pressure drop
volume constraint 15%



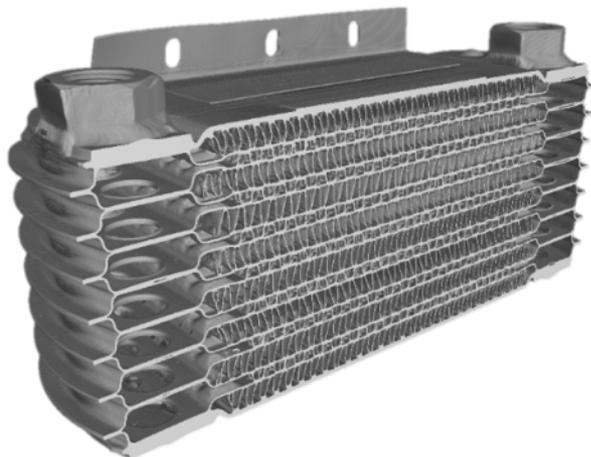
Two-Phase Heat Exchanger



Electrical Aircraft – X57 Maxwell

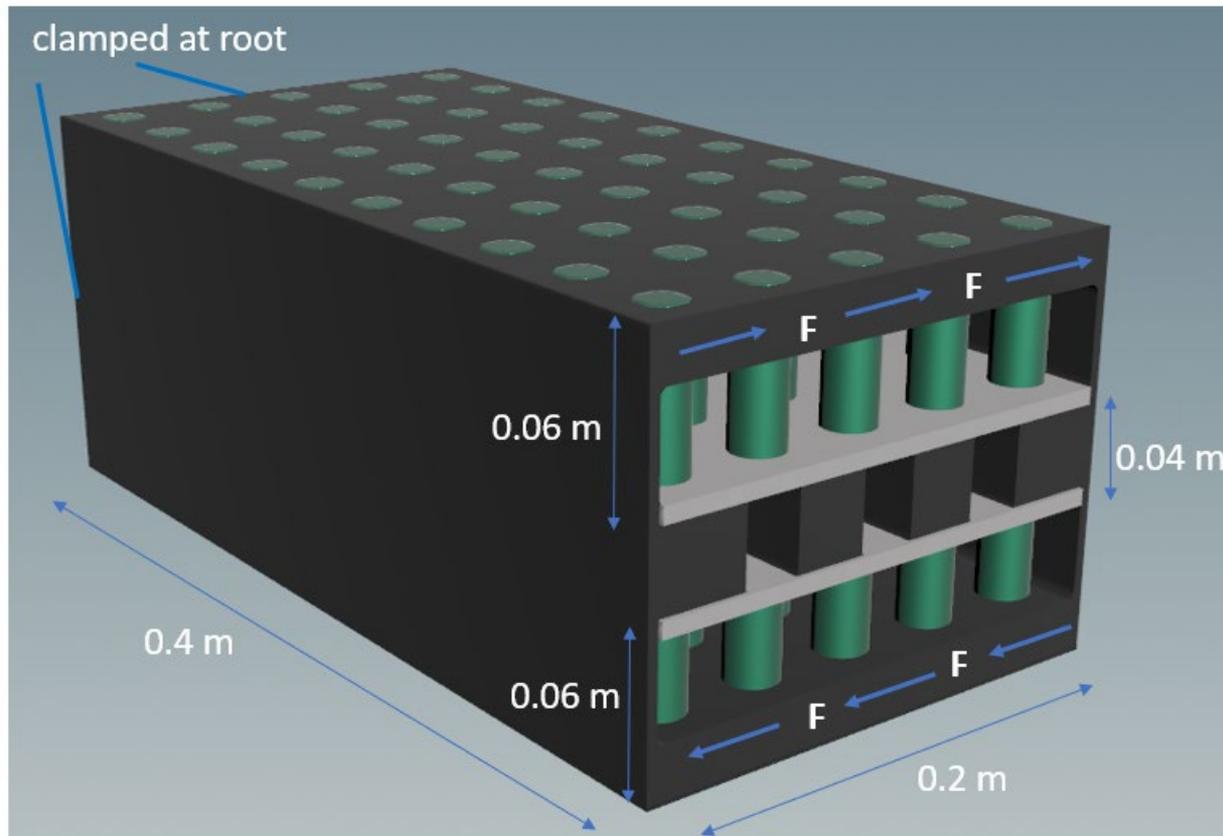


<https://www.antilehikoinen.fi/technology/electric-aircraft/>

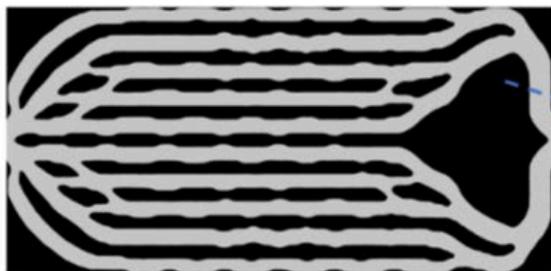


<https://sacd.larc.nasa.gov/x57maxwell/>

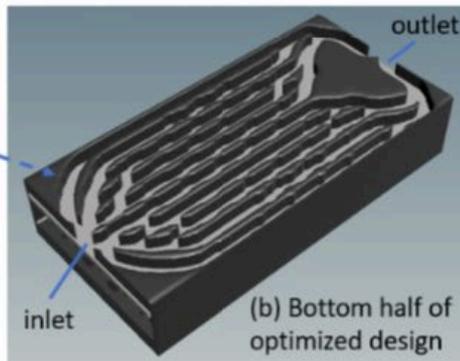
Load-carrying + Heat Exchanger + Battery Pack



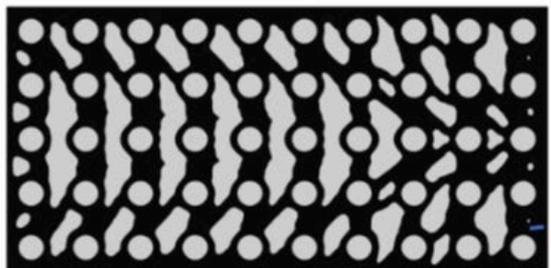
Load-carrying + Heat Exchanger + Battery Pack



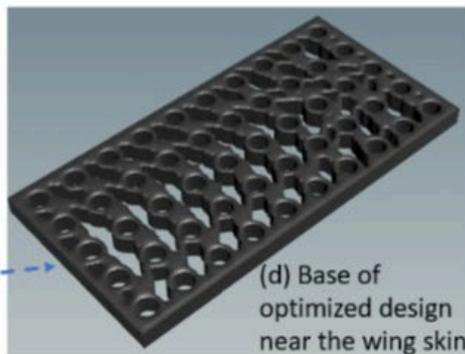
(a) Topology of cooling channels (Ω_2)



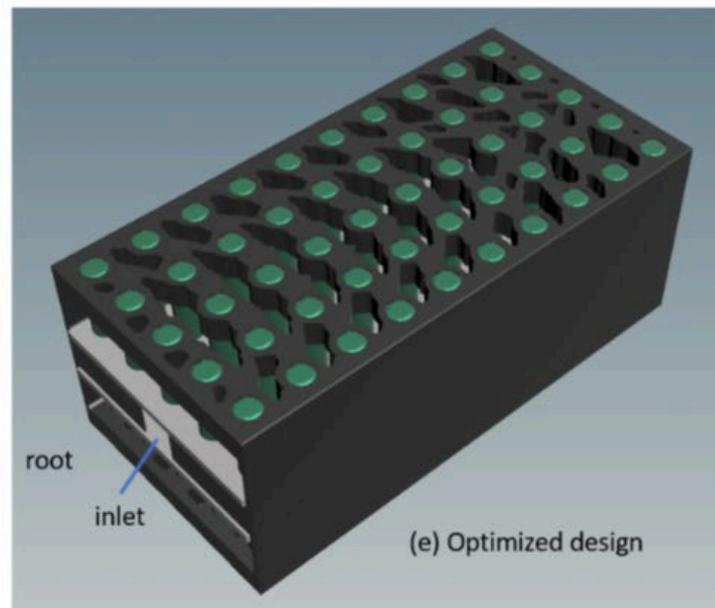
(b) Bottom half of optimized design



(c) Topology of structure at the wing skin (Ω_1)



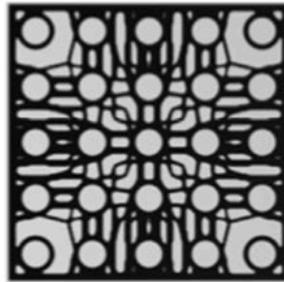
(d) Base of optimized design near the wing skin



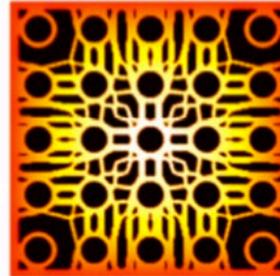
(e) Optimized design

Heat Conduction + Stress

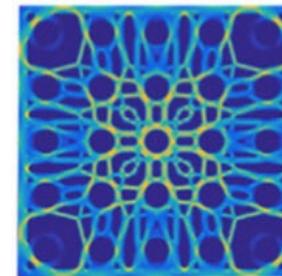
Minimize mass s.t.
 Stress \leq 100 MPa
 Max Temp \leq 50 °C



Mass = 2.51 kg



Max Temp = 50 °C

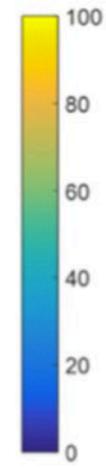


Max Stress = 100 MPa

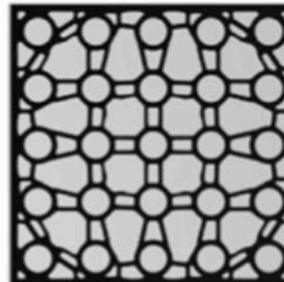
°C



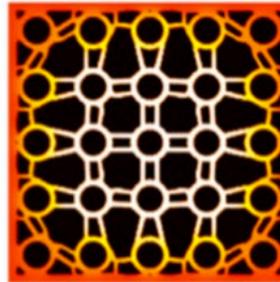
MPa



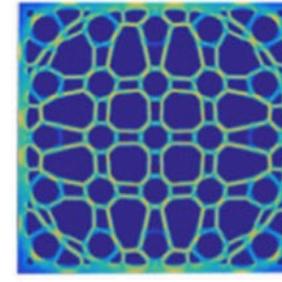
Minimize mass s.t.
 Stress \leq 100 MPa



Mass = 1.85 kg

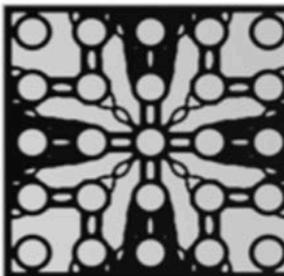


Max Temp = 81 °C

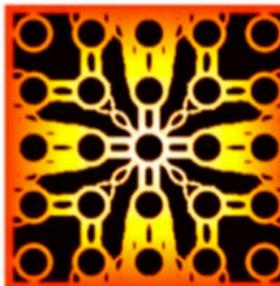


Max Stress = 100 MPa

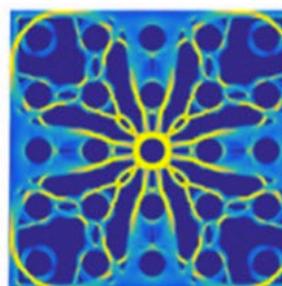
Minimize mass s.t.
 T \leq 50 °C



Mass = 2.38 kg



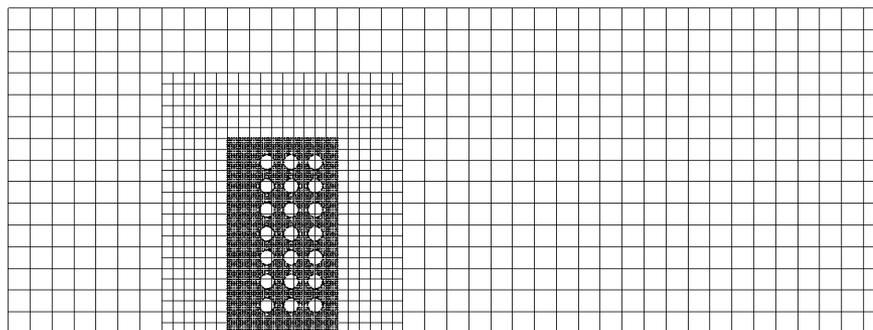
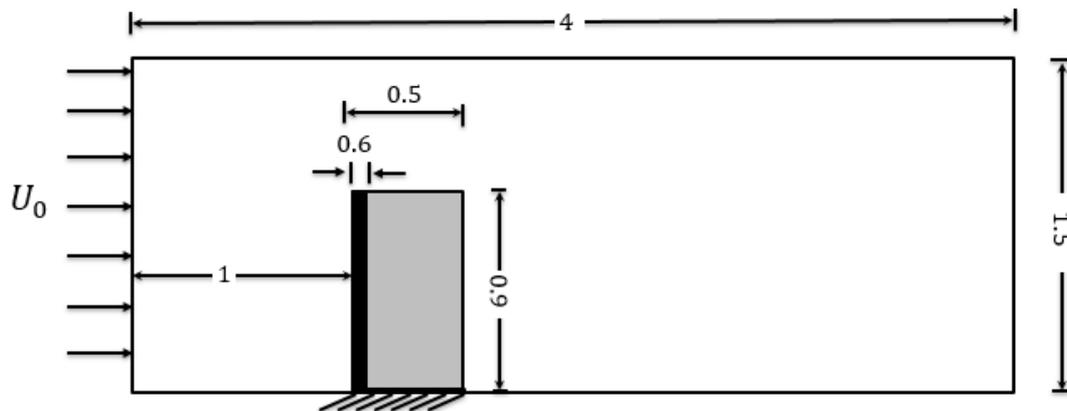
Max Temp = 50 °C



Max Stress = 273 MPa

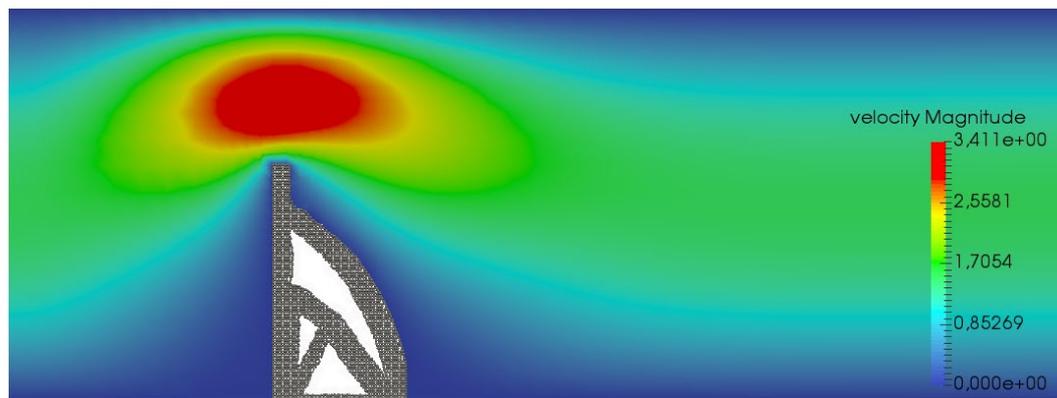
Leaflet in a Channel - FSI

- Steady-state flow
- Linear elastic solid
- Optimization:
Min. Compliance
s.t. Volume constraint 40%

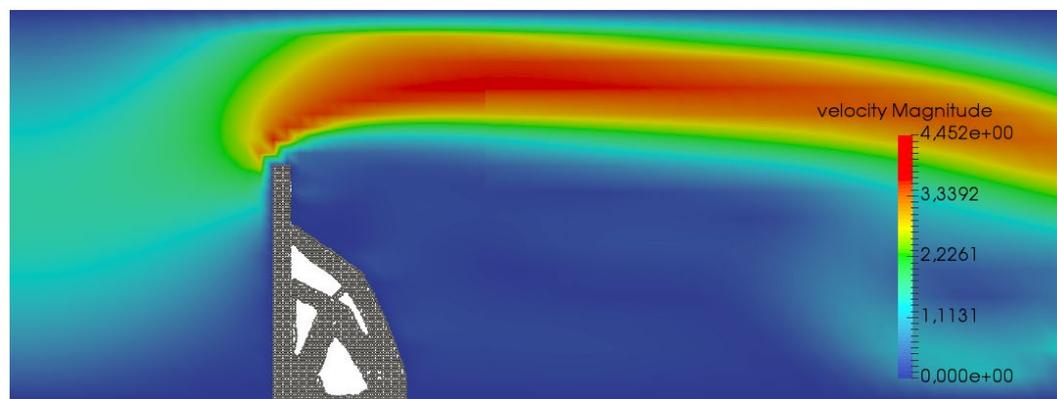


Solutions for Different Re

Re = 0.01



Re = 1000

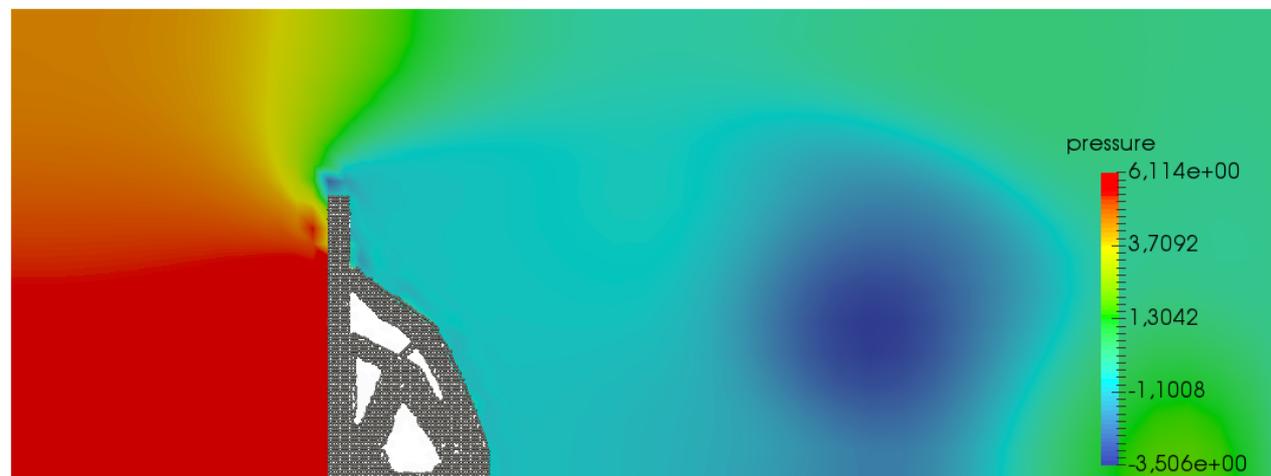


Pressure Profiles

Re = 0.01



Re = 1000





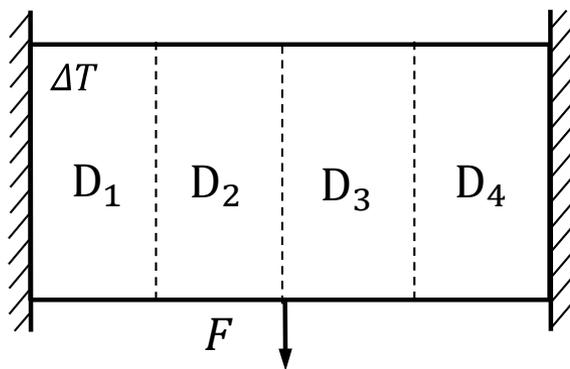
Multiscale Topology Optimization

ΔT (°C)	Macroscopic design	Microscopic design		Effective elasticity matrix (GPa)
		Unit cell	3x3 arrays	
0				$\begin{bmatrix} 219.23 & 65.77 & 0 \\ 65.77 & 219.23 & 0 \\ 0 & 0 & 76.73 \end{bmatrix}$
75				$\begin{bmatrix} 97.32 & -0.17 & 22.38 \\ -0.17 & 27.11 & 1.63 \\ 22.38 & 1.63 & 17.02 \end{bmatrix}$
225				$\begin{bmatrix} 56.75 & -7.32 & 6.85 \\ -7.32 & 9.73 & -0.79 \\ 6.85 & -0.79 & 6.15 \end{bmatrix}$
300				$\begin{bmatrix} 26.17 & -3.96 & 2.74 \\ -3.96 & 4.66 & -0.37 \\ 2.74 & -0.37 & 2.83 \end{bmatrix}$

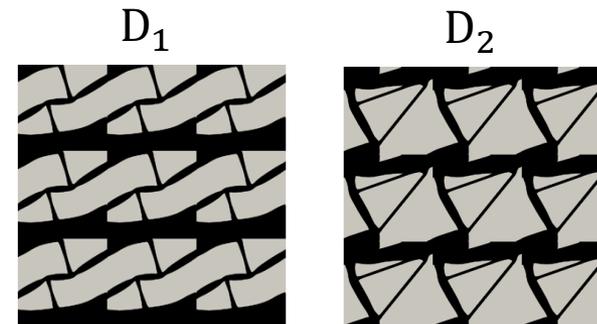
Multiscale Topology Optimization with Multiple materials



- 4 subdomains

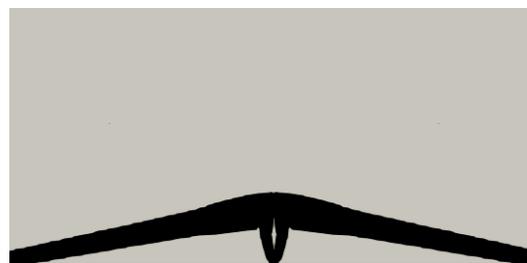
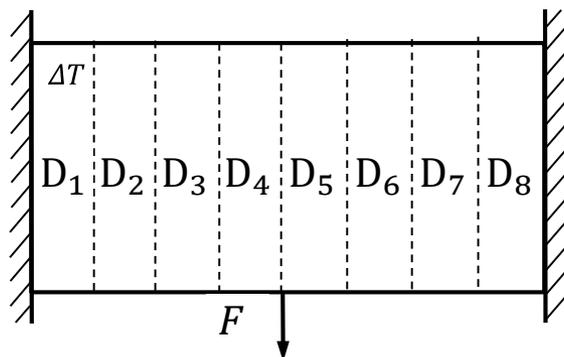


Macroscopic design
 $f = 8891.31, V_{mac} = 0.128$

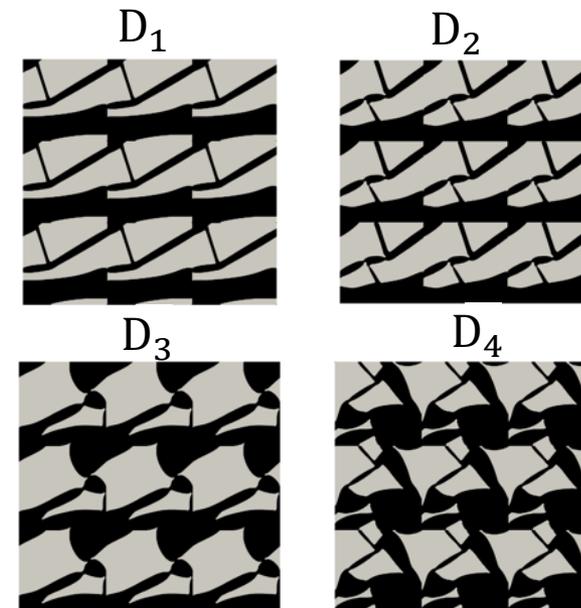


$V_{mic} = 0.391$

- 8 subdomains



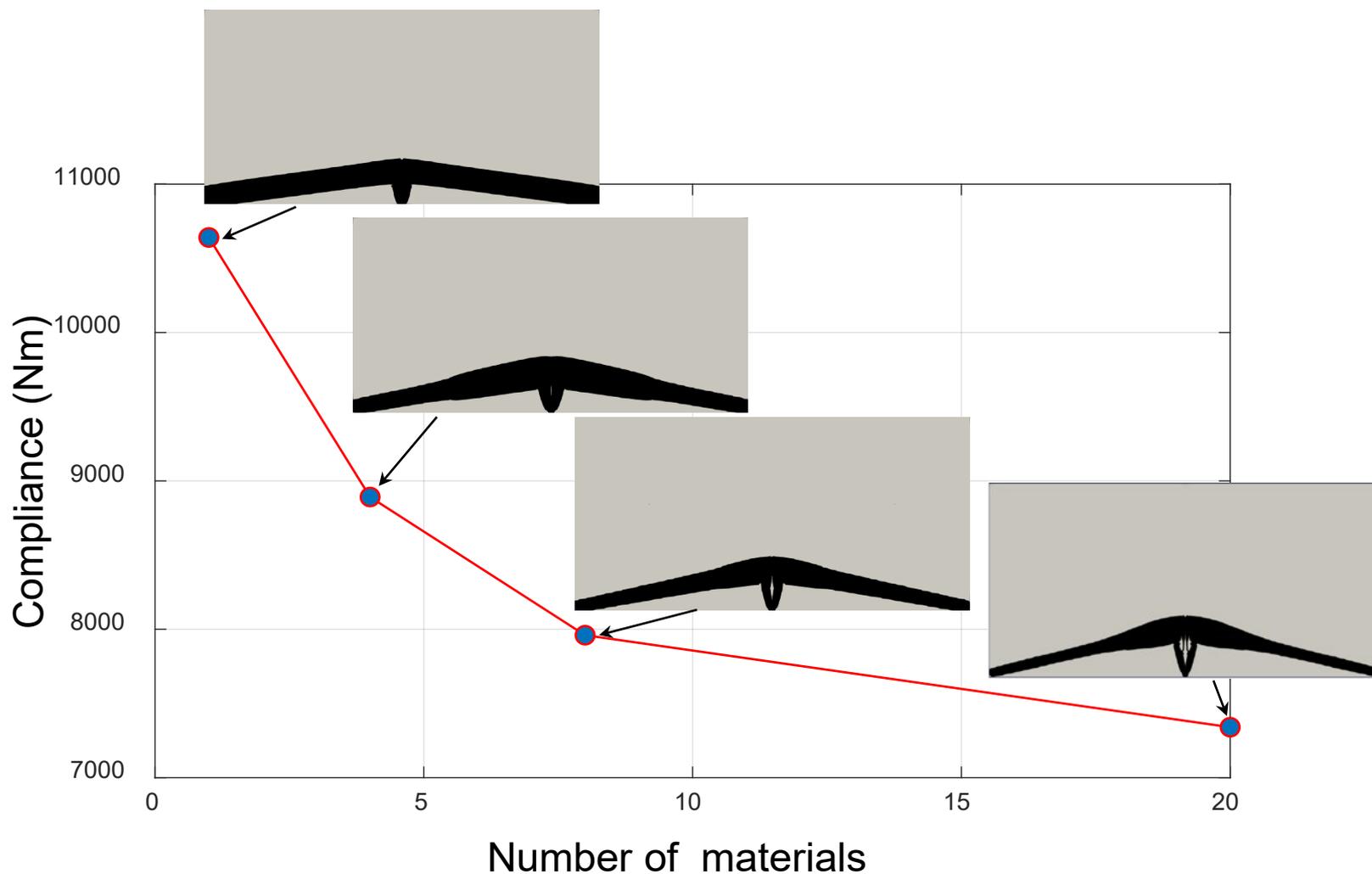
Macroscopic design
 $f = 7959.96, V_{mac} = 0.103$



$V_{mic} = 0.487$



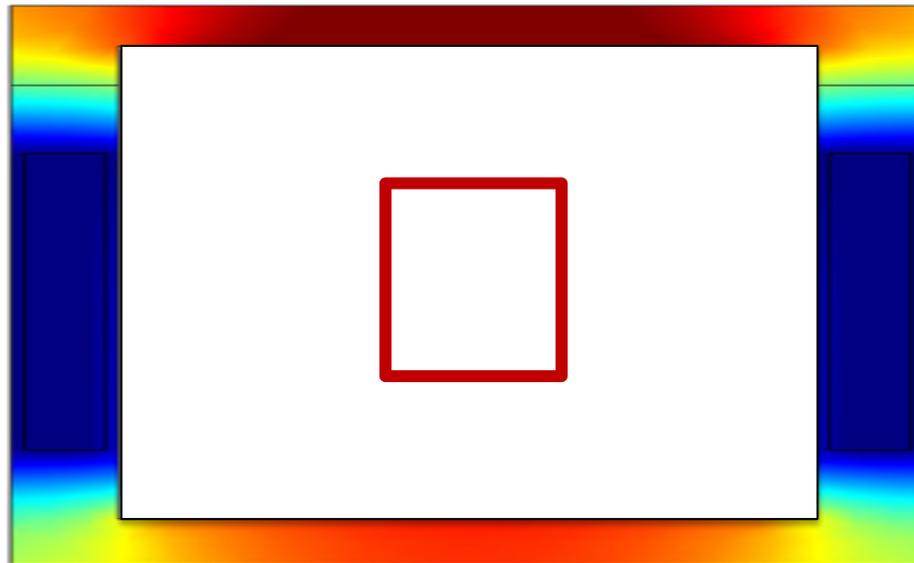
Pareto optima for multiscale solutions





Current Research: TO for mixed heat transfer modes

Temperature control under Conduction + Convection + Radiation





Optimization of Rover Chassis : Alex Guibert Introduction

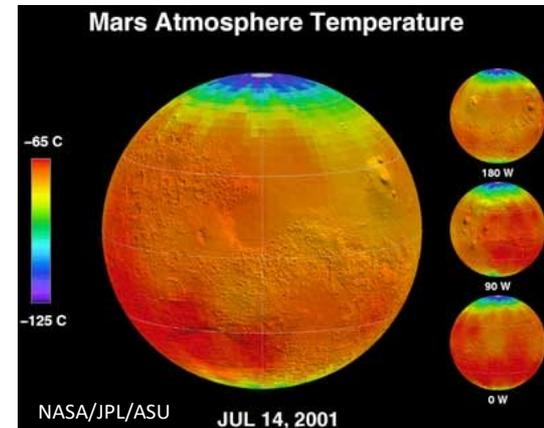
- R&TD Project
- Additively Manufactured Rover Chassis with Integrated Thermal Control for Extreme Cold Environments
 - Thermo-mechanical topology optimization of the chassis
 - Additively Manufactured heat switch
 - Additively Manufactured structural insulation



Motivations

- Develop a rover for extreme cold environment
- Cold Lunar regions / Lunar permanently shadowed regions (PSR) $\sim -230\text{ }^{\circ}\text{C}/50\text{ K}$
- Martian high latitudes $\sim -150\text{ }^{\circ}\text{C}/120\text{ K}$
- These environments are very challenging for the avionics: operational temperature $\sim -20\text{ }^{\circ}\text{C}/250\text{ K}$
- Active heating is needed to maintain the temperature

Objective: minimizing the amount of energy needed while satisfying the structural requirements

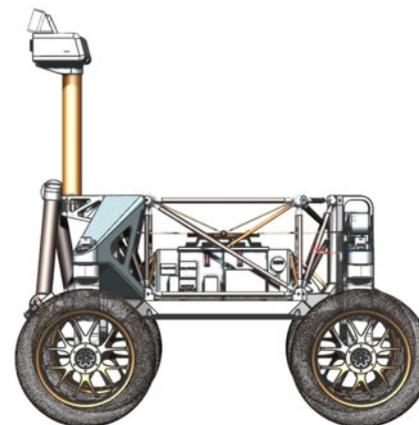
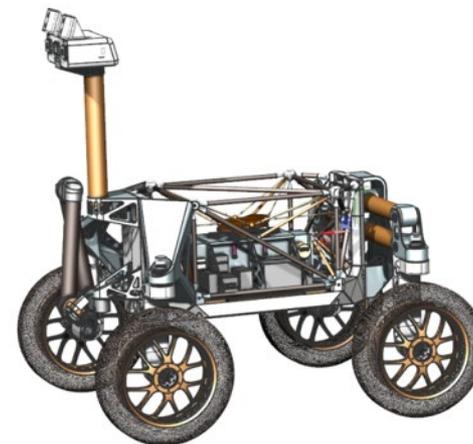
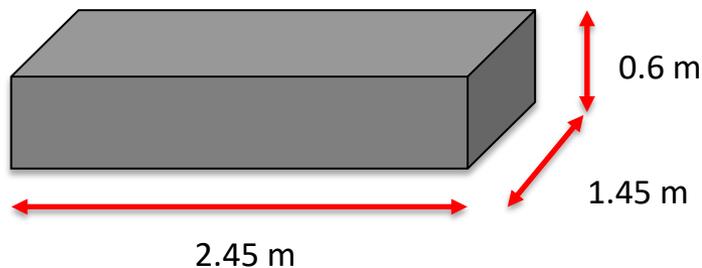


	Daytime High			Nighttime Low		
	$^{\circ}\text{C}$	$^{\circ}\text{K}$	$^{\circ}\text{F}$	$^{\circ}\text{C}$	$^{\circ}\text{K}$	$^{\circ}\text{F}$
Mean Surface	107	380	225	-153	120	-243
Equator (0° Latitude)	122	395	252	-158	115	-252
Mid-Latitudes	77	350	171	-143	130	-225
Poles	-43	230	-45	-63	210	-81
Dark Polar Crater	-233	40	-387	-233	40	-387

Lunar surface temperatures based on location, Heiken, 1991

Baseline

- Intrepid planetary mission concept study from 2020
- Design domain and loading conditions
- Mass of the mast ~ 120 kg
- Mass of batteries + electronics ~ 230 kg

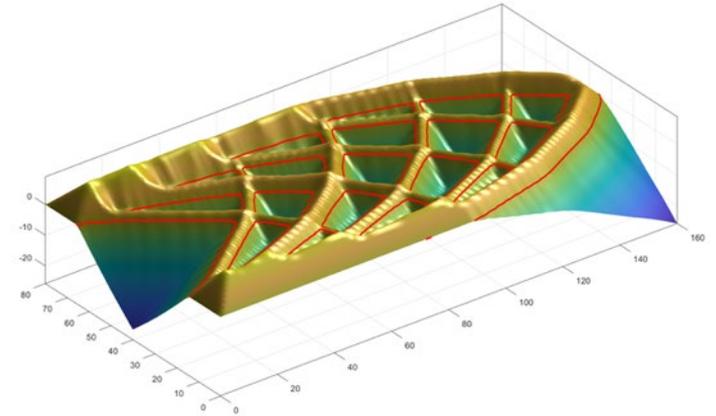


Intrepid Planetary Mission Concept Study Report, JPL/ASU, 2020



Approach

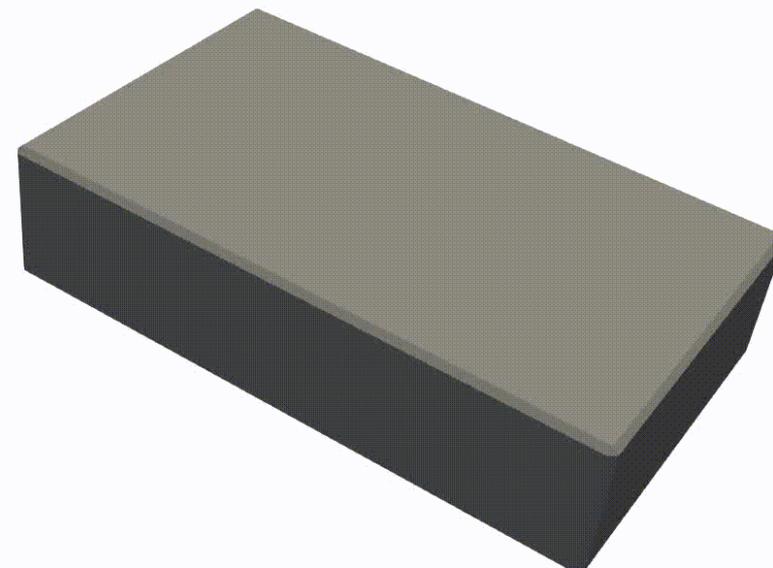
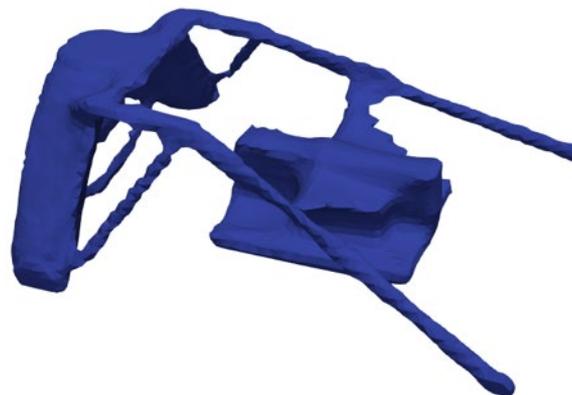
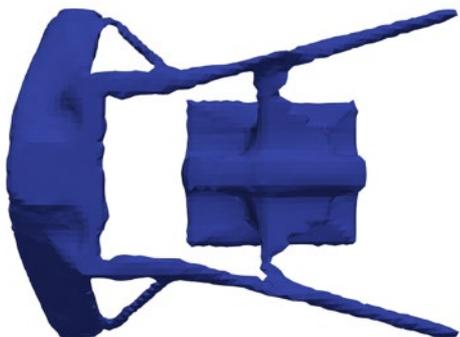
- Coupling of the thermal and structural disciplines to maximize the overall performance
- Level-set topology optimization
- Efficient implementation for large scale multiphysics optimization (parallel computing, low-level programming language for backend computation...)
- Experimental validation





Preliminary results

Maximization of the stiffness
Subject to a volume requirement



Titanium
Mass \sim 300 kg



Next steps

- Thermo-mechanical topology optimization of the rover chassis
- Validation through prototypes
- More complex loading conditions: launch, uneven surfaces...





Conclusions

- Conflicting multiphysics environments can create unintuitive design space.
- Topology Optimization can consider coupled multiphysics effects, leading to the overall optimum design.



